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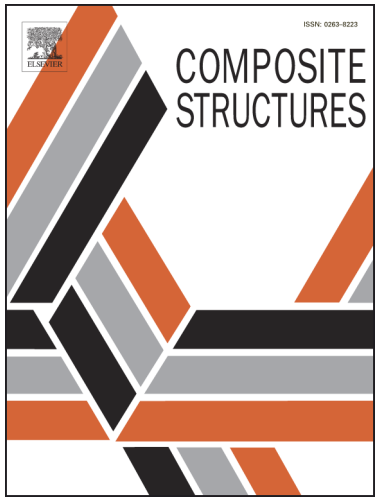

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Cohesive Zone and Level Set Method for Simulation of High Cycle Fatigue Delamination in Composite Materials

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Abstract

This paper deals with high cycle fatigue delamination in composite materials. The cohesive zone approach along with the level set method is used to simulate fatigue-driven delamination growth. The cohesive zone method is used for calculation of the energy release rate at the crack front because of its superiority over the virtual crack closure technique (VCCT) for bi-material interfaces and non self-similar crack growth. Evolution of the crack front in 3D during fatigue growth is handled with the level set method. The damage variable in the cohesive zone formulation is changed according to the updated level set field. Benchmarks are used to evaluate the performance of the proposed approach in simulation of 3D delamination growth under fatigue loading.

Keywords: Cohesive zone method, Level set method, Fatigue, Delamination, Fracture mechanics

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1. Introduction

Delamination is the separation of plies and is one of the most detrimental modes of failure in the composite materials. Delamination can be initiated by cyclic loading, impact, stresses near free edges, manufacturing defects like incomplete wetting or the presence of transverse matrix cracks. Delamination can grow under fatigue loading and lead to reduction of stiffness, lowering of critical buckling load and complete failure of the structure. Accurate modeling of delamination growth under fatigue loading is essential to safe design of composite structures exposed to cyclic loading such as wind turbine blades and aircraft wings.

For the case of small-scale yielding at the crack front, Paris' equation has been successfully used to describe crack growth under cyclic loading. First proposed by Paris et al. [1, 2], this equation is usually presented in terms of either stress intensity factor or energy release rate:

$$\frac{da}{dN} = C\left(\frac{\Delta G}{G_c}\right)^m \tag{1}$$

where C and m are material parameters which depend on the loading mode, G_c is the critical energy release rate which is also dependent on the mode ratio and ΔG is the cyclic range of energy release rate. The variable a signifies the crack length and N is the number of cycles. Although this equation was initially used for metals, it has been successfully applied to the laminated composites too [3, 4]. As it is shown in Eq. (1) the calculation of the range of energy release rate ΔG is required at the crack front. The virtual crack closure technique (VCCT) is often used in metals for calculation of this parameter. However for bi-material interfaces the assumption of linear elasticity leads to oscillatory singular fields at the

crack tip. This makes the application of fracture mechanics methods like VCCT far more complex [5]. Moreover VCCT is not very efficient for crack growth analysis, especially in cases with non self-similar crack growth since the elements need to be aligned with the crack front. Remeshing after a step of crack growth is usually required. This makes the simulation process time-consuming and complicates automation.

In this paper, the cohesive zone method is used to calculate the energy release rate. The cohesive zone concept was first introduced by Dugdale [6] and Barenblatt [7] and since then this method has been successfully used to model fracture in adhesive joints [8, 9] , bi-material interfaces [10, 11] and laminated composites [12, 13]. In the cohesive zone method, fracture is modeled by use of a nonlinear constitutive relation called the traction-separation law. This law provides a relation between the separation of two interfaces and the traction that opposes this separation. If the separation exceeds a limit called the final separation Δ^f, the traction will be reduced to zero and the crack advances. The traction-separation law is implemented in the finite element framework by use of the interface elements. A more detailed discussion of this model is presented in the next section.

The cohesive zone method has been mainly used for the simulation of crack growth under monotonic loading. Foulk et al. [14] were among the first researchers who extended the cohesive zone method for modeling fatigue crack growth. They achieved this by adding a unloading/reloading path to the Tvergaard's traction-separation law [15]. Several other fatigue models based on the cohesive zone concept have been proposed [16–19]. However these models are more suited for the cycle by cycle analysis and can result in high computational costs in high cycle fatigue simulations.

More recently alternative cohesive zone based models more suited for high cycle fatigue have been proposed by different researchers [20–22]. These models propose phenomenological relations for the growth of the damage parameter in the traction-separation law. As this damage parameter grows, the stiffness of the interface reduces. These damage growth models have some material parameters which need to be determined by means of experiments. Other researchers [23–25] have tried to link the damage growth to the Paris law. This link helps to avoid introduction of additional material parameters apart from the already well known Paris law parameters.

However, there is a difficulty in linking the damage mechanics from a cohesive law to the fracture mechanics of Paris law. In the cohesive law, the energy release rate is defined as the area under the complete traction separation curve. Fatigue cohesive laws have been proposed [23, 24], that accelerate damage development based on estimation of the energy release rate. However, during damage development, the final shape of the traction-separation curve and consequently, the energy release rate are yet unknown. Kawashita and Hallett [25] have presented an alternative approach where fatigue degradation is only applied in the element at the crack tip. Upon mesh refinement, this leads to a model where the local fatigue degradation is applied suddenly. A vertical drop in traction is approached. This implies that the complete area under the curve is known at the moment that fatigue damage is applied. For a 2D scenario, where there is a single crack tip element, this works very well. However, in 3D it involves a crack tip tracking algorithm and estimation of the direction of crack propagation.

As an alternative to cohesive zone modeling, Latifi et al. [26] have shown that using a level set approach is very suitable for fatigue analysis. Unlike cohesive laws

which deal with local damage development, the level set method deals with velocity of a front, which allows for a direct link with a crack growth rate. In [26], the model is a pure fracture mechanics approach without cohesive zone, but with dedicated element formulation. The model is limited to thin structures and so far only applicable to single delamination. The present paper combines ideas from Kawashita and Hallett [25] and the level set method. The level set method takes care of describing and updating the crack front location, while the cohesive law provides the estimate for the energy release rate. The cohesive law works exactly as the static cohesive law until the level set front passes the integration point. When the front passes an integration point, the damage is set to 1 in that point. This way, a true vertical drop in traction is achieved irrespective of element size. In the following sections first the basic formulation of the cohesive zone method is presented. Then the level set method and its implementation for triangular elements are reviewed and the proposed approach for simulation of delamination growth is discussed. Finally two benchmark cases are modeled to investigate the performance of the method both in calculation of the energy release rate and in tracking the crack front evolution in 3D.

2. Formulation

2.1. Cohesive Zone Method

The traction-separation law is implemented in the finite element framework by using the interface elements. Different constitutive formulations have been proposed [11, 27–34], among which the bi-linear traction-separation law is the most common one because of its simple shape. In this law an initial stiffness is introduced which ensures the stiff connection between interfaces before the damage initiation.

Therefore it contains an initial linear elastic part and a softening part. The shape of the traction-separation law is shown in Fig.1, where K is the initial stiffness of the interface, T^0 is the strength and G_c is the fracture energy.

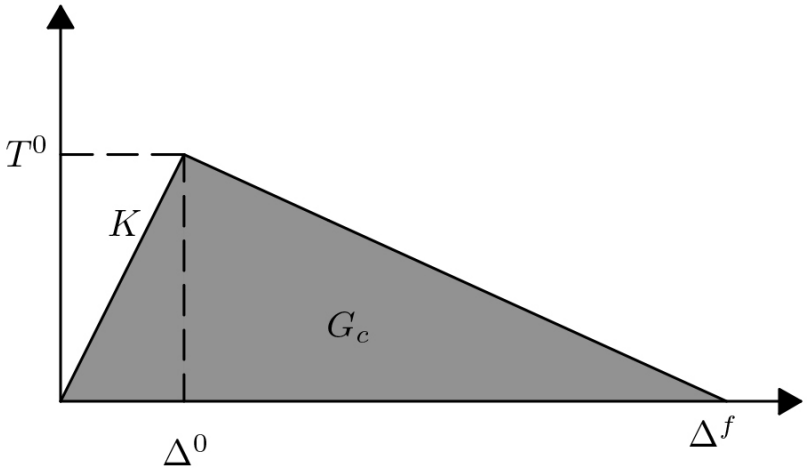

Figure 1: Bi-linear traction-separation law.

In this paper the quasi-static constitutive formulation presented by Turon et al. [28] is used as a starting point. A 3D element with two triangular faces and total number of 6 nodes is implemented in ABAQUS using the user-element subroutine UEL.

The bi-linear traction-separation law can be written as follows:

$$\begin{cases} T_i = K\Delta_i & \text{if} \quad 0 \leq \lambda \leq \Delta^0 \\ T_i = (1-d)K\Delta_i & \text{if} \quad \Delta^0 \leq \lambda \leq \Delta^f \\ T_i = 0 & \text{if} \quad \Delta^f \leq \lambda \end{cases} \tag{2}$$

where i shows the mode of loading, T is traction, Δ is the separation of interfaces and K is the penalty stiffness. In this equation λ, Δ^0, Δ^f and d are the equiv-

alent separation, initial separation, final separation and damage in mixed-mode respectively. The equivalent separation λ is the Euclidean norm of the separation components and is calculated as follows:

$$\lambda = \sqrt{\langle \Delta_I \rangle^2 + \Delta_{II}^2 + \Delta_{III}^2} \tag{3}$$

where $\langle\rangle$ is Macaulay brackets. The damage variable d has the following relation with other traction-separation law parameters:

$$d = \frac{\Delta^f(\lambda_{max}^\tau - \Delta^0)}{\lambda_{max}^\tau(\Delta^f - \Delta^0)} \tag{4}$$

where $\lambda_{max}^\tau = \max_{0 \le t \le \tau}\{\lambda(t)\}$ and τ is the time at which d is calculated.

The energy release rate is defined as the total work done per unit area for the complete failure of the material point. This value can be obtained by integrating the complete failure path in the traction-separation response. In the damage models proposed by Turon et al. [23] and Harper and Hallett [24], continuous increase of the damage variable under cyclic loading leads to gradual loss of stiffness in the interface. This increases the separation of the interface as the fatigue damage accumulates which leads to a non-vertical failure path (Fig. 2-a). The non-vertical failure path means the complete failure response and the total dissipated energy would be unknown at the moment t when the crack growth rate is being calculated. Kawashita and Hallett [25] limited the damage accumulation to the element adjacent to the crack tip. As a result the loss of stiffness would happen only to a small part of the interface and the failure path would be close to vertical. In the limit case where the element size at the crack tip goes toward zero, this path becomes completely vertical (Fig. 2-b). In this paper a vertical path is achieved

irrespective of the element size by setting the damage variable equal to 1 at nodes which have been passed by a moving front described with the level set method. At the integration points adjacent to the front, the energy release rate is known and can be obtained from the following relation:

$$G = \frac{T^0}{2}\left[\Delta^f - \frac{(\Delta^f - \lambda)^2}{\Delta^f - \Delta^0}\right] \tag{5}$$

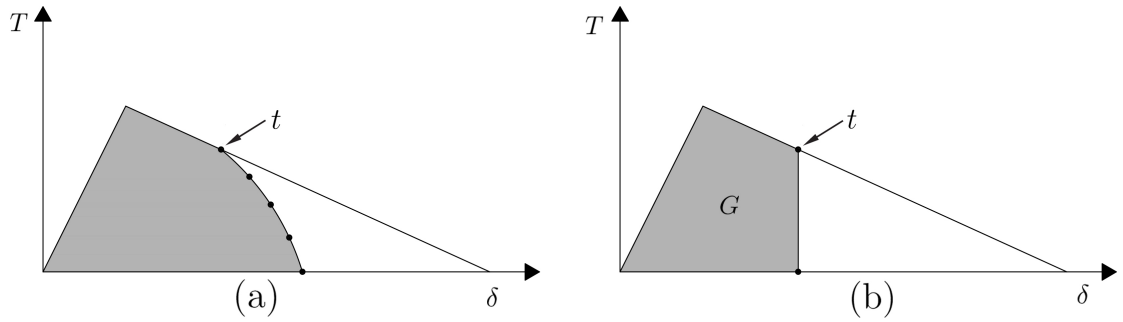

Figure 2: (a) Non-vertical failure path (b) Vertical failure path.

The cyclic variation of energy release rate ΔG in Eq. (1) is defined as:

$$\Delta G = G_{max} - G_{min} \tag{6}$$

By using the load ratio R, the cyclic variation of energy release rate ΔG can be computed by using G_{max}:

$$\Delta G = (1 - R^2)G_{max} \tag{7}$$

using Eq. (5), ΔG can be written as follows:

$$\Delta G = \frac{T^0}{2}\left[\Delta^f - \frac{(\Delta^f - \lambda_{max})^2}{\Delta^f - \Delta^0}\right](1 - R^2) \tag{8}$$

For the mixed-mode loading Δ^f and Δ^0 are found using the following relations [28]:

$$\Delta^f = \frac{\Delta_I^0 \Delta_I^f \left(\Delta_{shear}^0 \Delta_{shear}^f - \Delta_I^0 \Delta_I^f\right) \beta^\eta}{\Delta^0} \tag{9}$$

$$\Delta_0 = \sqrt{\left(\Delta_I^0\right)^2 \left(\left(\Delta_{shear}^0\right)^2 - \left(\Delta_I^0\right)^2\right) \beta^\eta} \tag{10}$$

where $\Delta_{shear} = \sqrt{\Delta_{II}^2 + \Delta_{III}^2}$ and $G_{shear} = G_{II} + G_{III}$. The parameter η is a material constant which is found by experiment. The variable β is defined as:

$$\beta = \frac{G_{shear}}{G_I + G_{shear}} \tag{11}$$

The fracture energy G_c in mixed-mode loading is described with the relation proposed by Benzeggagh and Kenane [35]:

$$G_c = G_{Ic} + (G_{II,c} - G_{Ic})\beta^\eta \tag{12}$$

where subscript c is used to show critical energy release rate.

2.2. Level Set Method

The level set method is a robust tool to track the evolution of moving fronts [36]. In this method the front location is represented by a level set function and as the front moves this function evolves in time. Knowing the velocity at different points of the front, a differential equation for the level set function evolution can be formed [37] :

$$\frac{\partial \varphi}{\partial t} + \boldsymbol{\nabla}\varphi \cdot \boldsymbol{V} = 0 \tag{13}$$

where φ is the level set function and $\boldsymbol{V}$ shows the velocity field. It is beneficial to use a signed distance function as the level set function. The absolute value of the signed distance function at each point shows the shortest distance of that point to the crack front and its sign shows on which side of the crack front the point is located. In the proposed method, the front that is tracked is the crack front that separates the partially damaged cohesive zone from the traction free crack. The positive sign is assigned to the uncracked side and the negative sign to the cracked side (Fig. 3). When φ is a signed distance function, $\boldsymbol{\nabla}\varphi$ is a unit vector and therefore:

$$\boldsymbol{\nabla}\varphi \cdot \boldsymbol{V} = V_n \tag{14}$$

where V_n is velocity value normal to the level sets of φ such as crack front. Using forward Euler time discretization, the level set function φ after time Δt can be found from the following relation [38]:

$$\varphi_{t+\Delta t} = \varphi_t - V_n \Delta t \tag{15}$$

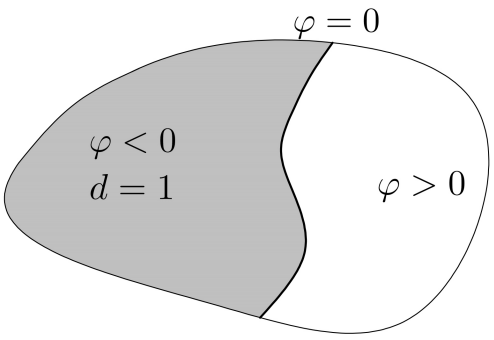

Figure 3: Signed distance function and damage values in cracked domain.

As was discussed in the previous section the energy release rate is calculated at the nodes on the crack front by measuring the area under the traction-separation curves. Since the level set method is used, the crack front is not necessarily aligned with the element boundaries and may pass through the elements. In this case, as shown in Fig. 4, the integration of the traction-separation law will be performed at the nodes of the elements cut by the crack front. A Newton-Cotes integration scheme is used for the cohesive elements which means integration points and nodes are coincident. Therefore values of the energy release rate can be readily calculated at nodes. After energy release rate values are obtained, crack growth velocities at these nodes are found using Paris equation (Eq. (1)). The crack growth rate da/dN is used as the velocity V_n required in the level set update equation (Eq. (15)). The time increment Δt in Eq. (15) is actually the number of cycles ΔN in fatigue crack growth. This discretization in time in fatigue problems is called the cycle jump. The cycle jump for any individual time step is chosen using the following equation [23]:

$$\Delta N = \frac{\Delta a_{max}}{\max\{\frac{da}{dN}\}} \tag{16}$$

where Δa_{max} is maximum amount of crack growth per time step and is pre-established, while $\max\{\frac{da}{dN}\}$ is the maximum value of the crack growth rate computed along the crack front. Reducing the value of Δa_{max} leads to more accurate results.

For updating the level set function φ, the velocity V_n should be known throughout the domain. Here the fast marching method [37] is used to extend the velocities from the crack front over the whole domain of the interface. The method proposed in [39] is used for extending velocities in triangular elements. In the ele-

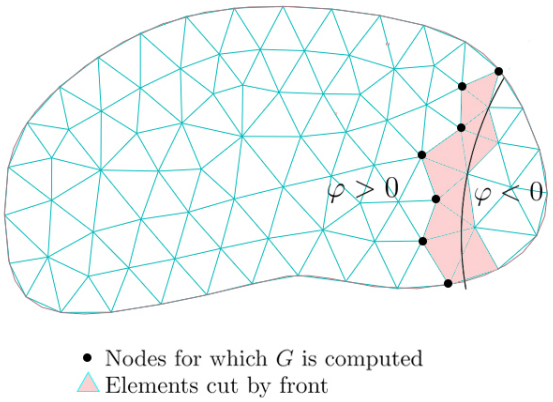

Figure 4: Calculation of energy release rate at the crack front.

ments where velocities at two nodes are known, the velocity at the other node is found by solving the normality condition which states that the velocity is constant in the direction normal to the φ level curves:

$$\boldsymbol{\nabla}\varphi \cdot \boldsymbol{\nabla} V_n = 0 \tag{17}$$

If a node is connected to more than one element with known velocities at two nodes, the element which is the most normal to the level set will be used. This is the element with the highest value of $|\nabla\varphi \cdot \nabla N_i|$ where i shows the node with unknown velocity.

While ideally $\varphi_{t+\Delta t}$ will be obtained as a signed distance function, approximation introduced in numerical solution of Eq. (13) will cause deviations. To ensure that the obtained level set function at the next time step $\varphi_{t+\Delta t}$ is a signed-distance function, reinitialization is performed. The reinitialized level set field will have the same zero level curve as $\varphi_{t+\Delta t}$. Reinitialization is performed by solving the equation $|\nabla\varphi| = 1$ by the fast marching method. On a triangular element with two known nodal values, this is a quadratic equation and the root with maximum

absolute value will be chosen. If a node is connected to multiple elements, the element with minimum value for $\max\{|\varphi_j|, |\varphi_k|\}$ will be chosen, where j and k are the nodes with known values of φ.

After $\varphi_{t+\Delta t}$ is found, the damage values at the nodes are calculated. For the nodes where $\varphi_{t+\Delta t} < 0$ damage values are set to 1 (Fig. 5). These damage values will be read by the UEL subroutine and therefore the new cracked region will be introduced to the finite element model. The finite element model will be solved again in the next time step with the new crack front and the above described steps will be repeated. The steps of the described procedure are shown in Fig. 6. ABAQUS is used for solving the finite element model. The other tasks in the procedure are performed with a Python code.

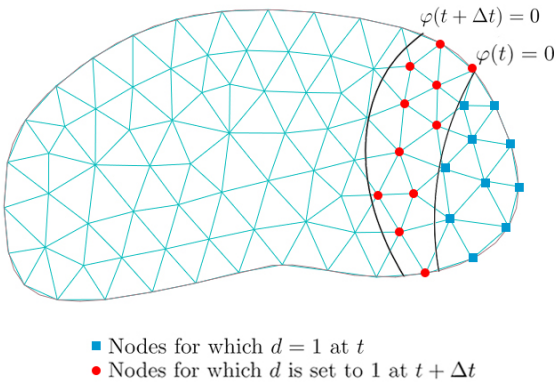

Figure 5: Change of the damage values at the nodes as the front moves.

3. Results and Discussion

In this section a 3D double cantilever specimen and a circular delamination specimen are used to evaluate the performance of the presented approach to simulate fatigue-driven delamination. These two cases are loaded in mode I and mode II

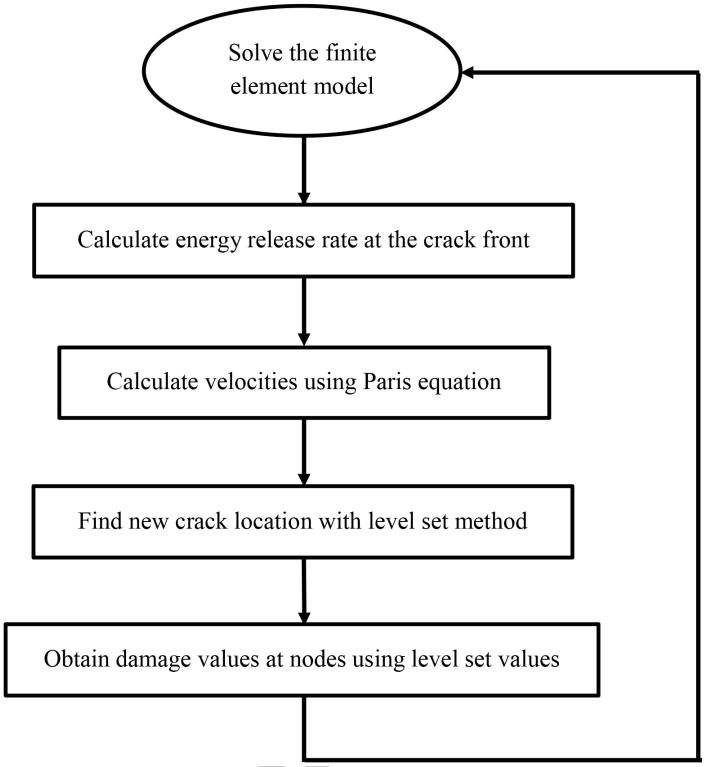

Figure 6: The steps of crack growth procedure.

respectively. The ability of the method to calculate the energy release rate accurately and to track the crack front is tested.

3.1. 3D Double Cantilever Beam specimen

To investigate the ability of the model to calculate the energy release rate and track the crack front in 3D problems, a double cantilever beam specimen is modeled using cohesive elements. The deformed finite element model of the specimen is shown in Fig. 7 with amplified deformations. The specimen has a total length of 20 mm with a delamination length of 10 mm. The specimen width is 10 mm and each arm has a thickness of 0.5 mm. Moments of equal values and opposite directions are applied to each arm while the other end of the specimen is completely fixed. The material properties for HTA/6376C carbon/epoxy are taken

from [23, 40, 41] and shown in Table 1. The laminate is considered to be unidirectional where the fiber orientation is aligned with the beam axis.

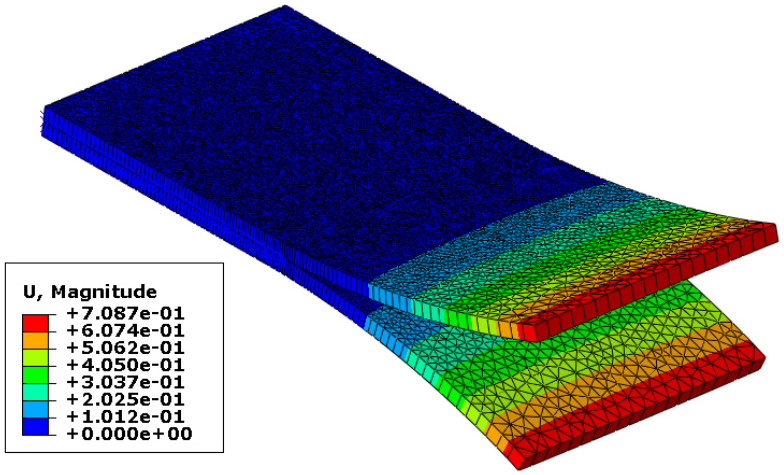

Figure 7: 3D double cantilever beam specimen.

The finite element model as shown in Fig. 7 is made from ABAQUS continuum shell wedge elements (SC6R) and the user-defined cohesive elements. Cohesive elements with zero thickness are placed in the delamination plane and a fine mesh with the element size of 0.1 mm is used ahead of the crack front to ensure that the traction distribution in the cohesive zone is captured with good accuracy. The cohesive zone is found as the region ahead of the crack front where damage d has a nonzero value and material points are in the softening zone (Fig. 1).

For the minimum number of elements in the cohesive zone different values have been suggested in the literature. Moës and Belytschko [42] suggested the minimum number of 10 elements in the cohesive zone, while Dávila and Camanho [43] have used 3 elements in their analysis. Because of the small length of the cohesive zone, even using 3 elements in the cohesive zone requires a small element size. Turon et al. [44] have suggested the lowering of the interfacial strength value

T^0 to increase the length of the cohesive zone and allow for the bigger elements. However as shown in Fig. 8 too much lowering of the interfacial strength creates problems in capturing the high gradient of G near the free edge. Because of the symmetry the energy release rate distribution on only one half of the front is plotted. In what follows the value of 20 MPa is used for interfacial strength to ensure enough elements exist in the cohesive zone. With this value, the cohesive zone spans approximately 8 elements.

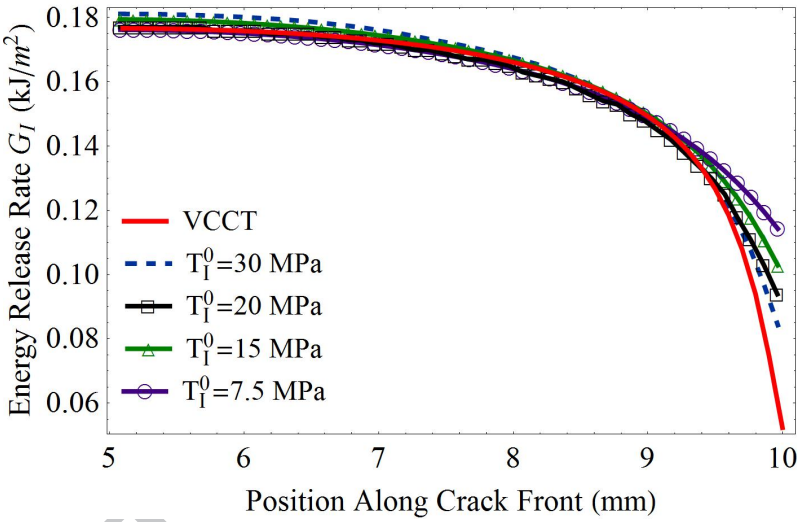

Figure 8: Energy release rate distribution along initial crack front for DCB specimen for different values of interfacial strength.

Table 1: Material properties of HTA/6376C carbon/epoxy [23, 40, 41].

Elastic Constants		Interface Properties		Paris Law Constants	
E_{11} (GPa)	120	G_{Ic} (kJ/m^2)	0.260	C_I (mm/cycle)	0.0616
$E_{22} = E_{33}$ (GPa)	10.5	G_{IIc} (kJ/m^2)	1.002	m_I	5.4
$G_{12} = G_{13}$ (GPa)	5.25	K (N/mm^3)	10^6	C_{II} (mm/cycle)	2.99
G_{23} (GPa)	3.48	T_I^0 (MPa)	30	m_{II}	4.5
$\nu_{12} = \nu_{13}$	0.3	T_{II}^0 (MPa)	60		
ν_{23}	0.51	η	2.73		

The crack front evolution for 15 steps with $\Delta a_{max} = 0.2$ mm is demonstrated in Fig. 9. To make sure that the accuracy is in an acceptable range, a small value for the step size is chosen. Because of the drop in the energy release rate value near the free edges (Fig. 8) the initial straight line changes to a curved shape. As was discussed in section 2.2 when the level set function is updated and the new crack front is found, damage values at the integration points with the negative value of the level set function are set to 1 (Fig. 5). This will cause the crack front to fall inside the cut elements and be influenced by the shape of the element boundaries. As a consequence oscillations may occur in the energy release rate values computed at the nodes. These oscillations reduce by mesh refinement and can be removed by applying a smoothing procedure. Here the smoothing is applied each time the finite element model is solved by ABAQUS and the energy release rate is obtained at the nodes of elements that are cut by the front. Oscillations in the energy release rate will be discussed in more detail in the next example.

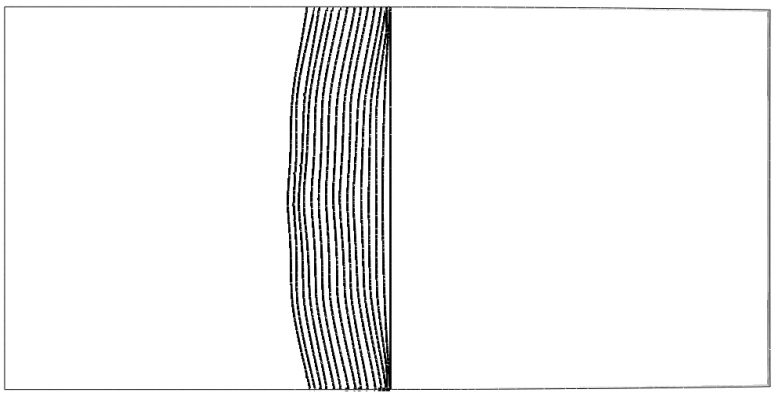

Figure 9: Evolution of crack front under cyclic loading for DCB specimen.

3.2. Circular Delamination Specimen

In this section another example is studied to investigate the performance of the presented approach. A circular delamination test with a central crack for mode II delamination is modeled [45]. The position of the circular crack and the boundary conditions are schematically shown in Fig. 10. With the depicted boundary conditions the crack growth will be in mode II only. The specimen has a radius of 20 mm, thickness of 1 mm and the radius of the initial crack is 5 mm. The material of the specimen is HTA/6376C carbon/epoxy (Table. 1). The laminate is made from two $[-45/90/45/0]_s$ sublaminates. The initial crack is located between the two sublaminates. This layup makes the laminate and the individual sublaminates quasi-isotropic which means that the in-plane material properties can be homogenized. The homogenized properties are shown in Table 2. Paris equation material parameters C and m for mode II are taken equal to 2.99 mm/cycle and 4.5 respectively.

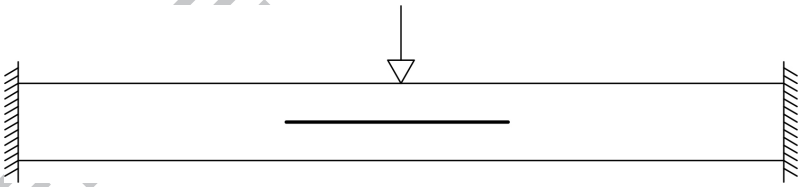

Figure 10: Circular Delamination Specimen.

Table 2: Homogenized material properties of HTA/6376C carbon/epoxy.

E (GPa)	ν	G_{IIc} (kJ/m^2)	T^0_{II} (MPa)
47.77	0.3	1.002	60

The finite element model constructed for the circular delamination test is shown in Fig. 11. The load applied to the center of the specimen is 400 N. Two rows of

the ABAQUS continuum shell wedge elements (SC6R) at above and below the delamination plane are used. These elements follow the first-order shear deformation theory and only have translational degrees of freedom. Between these two rows, one layer of cohesive elements with zero thickness is inserted. Although the model with homogenized laminate properties is axisymmetric and can be reduced to a 2D model, this will not be favorable since the goal here is to investigate 3D crack growth. However symmetry is used to only model a quarter of the specimen. To capture the traction distribution accurately at the crack front, a fine mesh is used. In the initial delaminated area which represents a circle with radius of 5 mm no cohesive elements are used and contact is defined to prevent interpenetration.

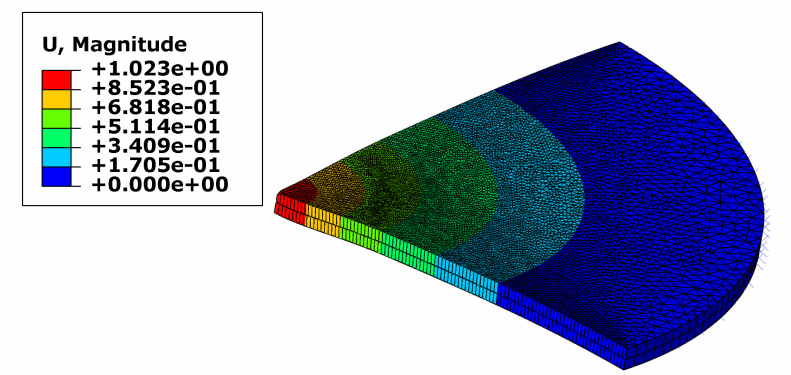

Figure 11: Finite element model of the circular delamination specimen.

As it is shown in the flowchart depicted in Fig. 6 the first step is to solve the finite element model and calculate the energy release rate at the crack front. The cohesive zone obtained from the finite element solution of the model in the first step is shown in Fig. 12.

As shown in Fig. 4, the crack front may pass through elements and is not necessarily aligned with the element boundaries. In the elements that are cut by the crack front the damage value d (Eq (4)) in the nodes on the cracked sides is set to

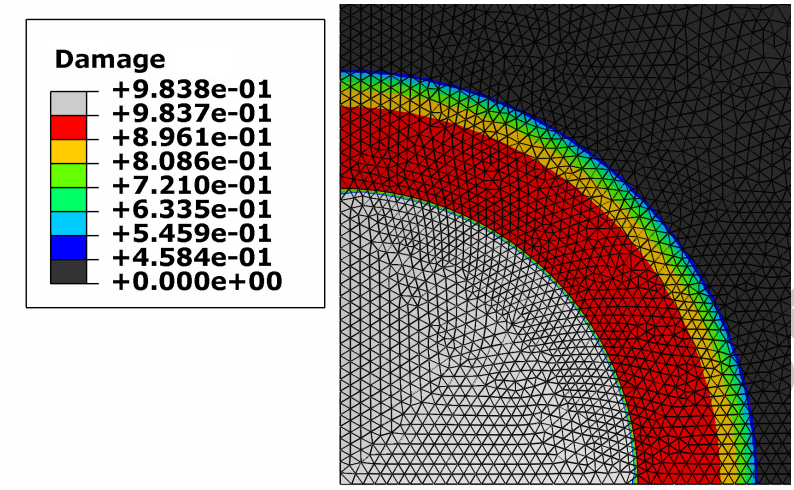

Figure 12: Cohesive zone ahead of the crack front.

1. This will cause the crack front to fall somewhere inside the element and be affected by the shape of the element boundaries. This may create oscillations in the calculated energy release rate values. However by reducing the element size the oscillations will reduce and the solution will improve. Energy release rate values for a circular crack with radius of 10 mm where the crack is no longer aligned with the element boundary are shown in Fig. 13 for two mesh sizes. The angle θ in Fig. 13 is the angle with the x axis which is aligned with the horizontal edge of the specimen in Fig. 12. As it is seen in this figure oscillations have reduced for the smaller mesh size. Besides reducing the element size, smoothing can also be used for alleviating the oscillations. This will be discussed in more detail in the next paragraphs. For obtaining the energy release rate by VCCT a 2D axisymmetric model is made and ABAQUS' own implementation of VCCT is used.

The energy release rate calculation and crack growth for an elliptical crack is also examined. An elliptical crack with major axes of 5 and 10 mm is inserted in the mid-plane of circular delamination specimen. As shown in Fig. 14 good agreement

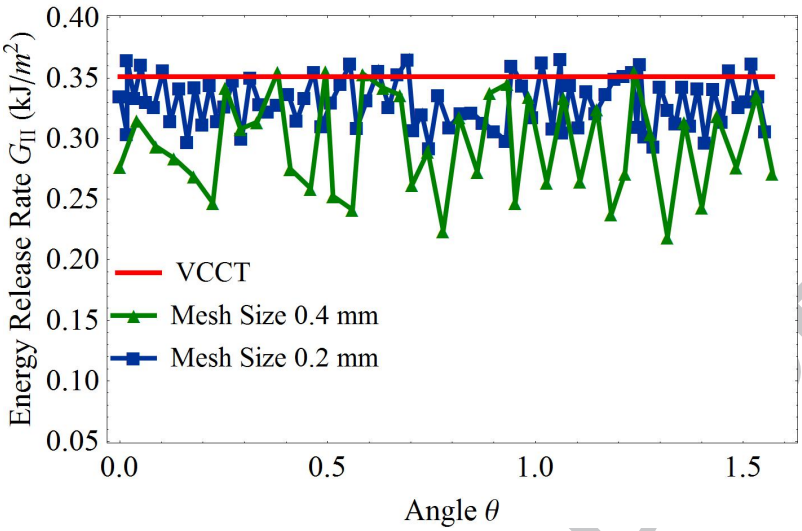

Figure 13: Energy release rate distribution along crack front.

exists between energy release rate values calculated using cohesive elements and values obtained from VCCT. Because for the initial crack, element boundaries are actually aligned with the crack front, the energy release rate curve is smooth and no oscillations are present.

As it is seen in Fig. 14 the maximum energy release rate value occurs at $\theta = 0$. The size of the cycle jump ΔN is obtained by choosing Δa_{max} equal to 0.3 mm (Eq. (16)). A small value for Δa_{max} is chosen to ensure that the results have acceptable accuracy. The damage values assigned to integration points after 3 steps of crack growth are depicted in Fig. 15. The white area in Fig. 15 shows the initial crack and the red area shows the crack extension. The finite element model is solved using these damage values and energy release rate along the new crack front is calculated. Obtained energy release rate values for this step are plotted in Fig. 16. As it is seen in this figure oscillations exist in the energy release rate values. As was discussed earlier the noise can be reduced by refining the mesh but also a smoothing procedure can be applied to remove the oscillations. The

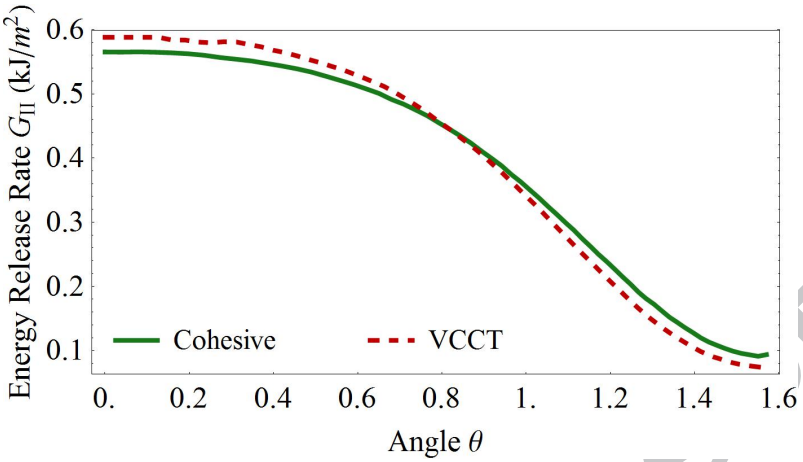

Figure 14: Energy release rate at crack front.

smoothed curved is plotted in Fig. 16. Here the Savitzky-Golay filter [46] has been used for smoothing.

As it is observed from Fig. 16 the energy release rate can be obtained accurately by integrating the traction-separation curve at integration points in elements cut by the crack front. The oscillations which occur in the energy release rate calculation can be alleviated by refining the mesh or by applying a smoothing procedure. The crack front evolution for 20 steps with $\Delta a_{max} = 0.3$ mm is demonstrated in Fig. 17. The initial elliptical crack is expected to grow to a circular shape. As it is seen in Fig. 17 the proposed method tracks the crack evolution to a circular front very well. A slight deviation from a true circle is observed which can be reduced by refining the mesh. As it is seen in Fig. 15 the meshing of the specimen is completely irregular. The presented benchmarks show the ability of the proposed approach to efficiently handle the two main constituents of fatigue crack growth simulations which are the calculation of energy release rates and the tracking of the crack front.

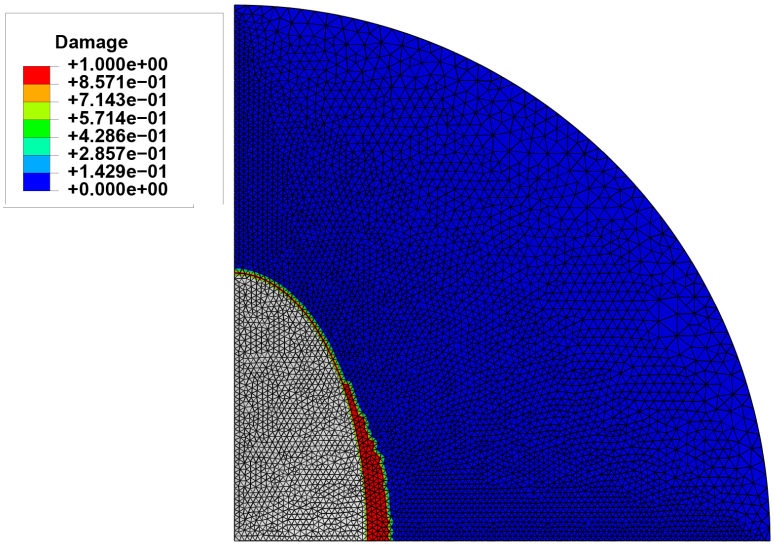

Figure 15: Damage values assigned to integration points.

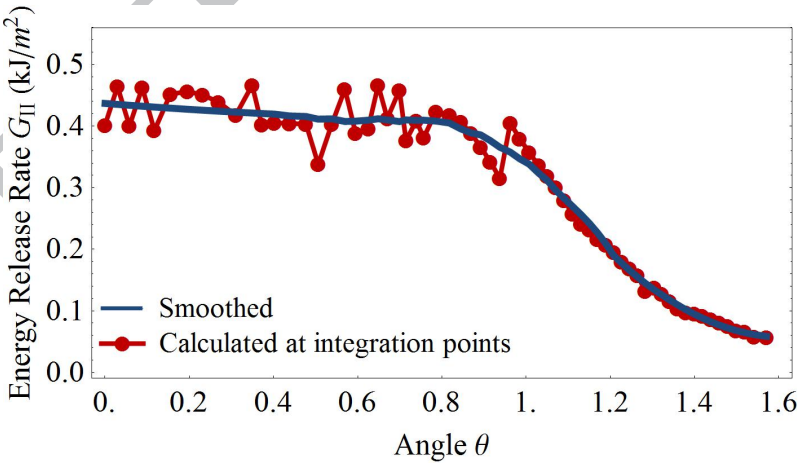

Figure 16: Energy release rate at crack front.

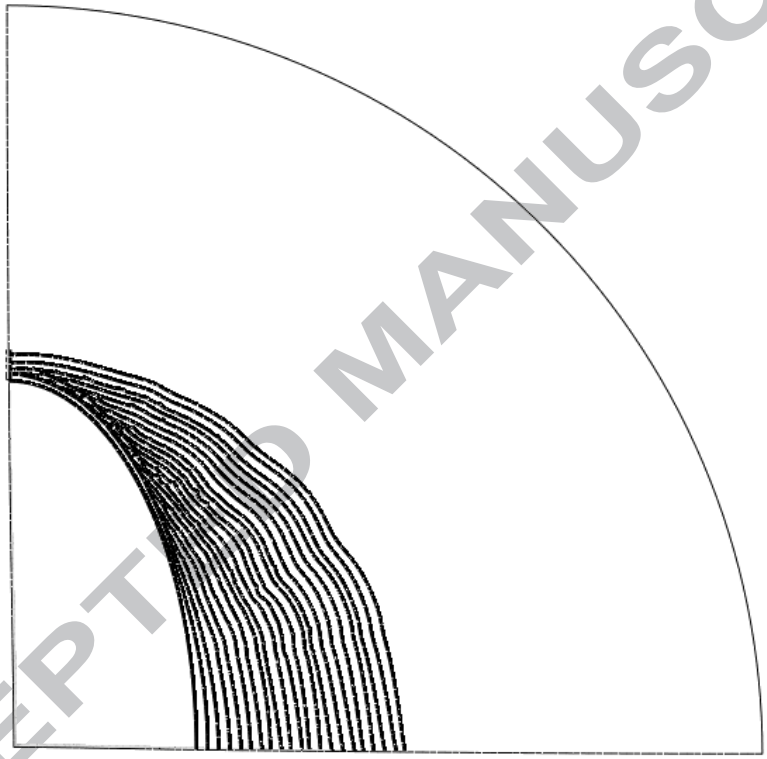

Figure 17: Crack front evolution.

4. Conclusion

A new method is proposed for modeling high cycle fatigue delamination in composite materials. The presented approach is based on a cohesive zone model and uses the level set method for tracking the crack front evolution in 3D problems. The integration of the traction-separation curve at integration points is used to calculate the energy release rate accurately. This method is superior to VCCT since it can be used for bi-material interfaces and no remeshing is required for arbitrary crack growth. Smoothing is proposed to reduce the oscillations of the calculated energy release rates at integration points. Since the velocity at the nodes is required by the level set method, the Newton-Cotes integration scheme is used. Unlike previous fatigue models based on the cohesive zone method, the presented approach does not introduce new material parameters or require calculation of an effective length. The benchmarks show the ability of the method to simulate fatigue-driven delamination and track the front evolution in 3D.

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